



# A Mission Concept to Determine the Magnetospheric Causes of Aurora

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Insufficiently accurate magnetic-field-line mapping between the aurora and the equatorial magnetosphere prevents us from determining the cause of many types of aurora. An important example is the longstanding question of how the magnetosphere drives low-latitude (growth-phase) auroral arcs: a large number of diverse generator mechanisms have been hypothesized but equatorial magnetospheric measurements cannot be unambiguously connected to arcs in the ionosphere, preventing the community from identifying the correct generator mechanisms. Here a mission concept is described to solve the magnetic-connection problem. From an equatorial instrumented spacecraft, a powerful energetic-electron beam is fired into the atmospheric loss cone resulting in an optical beam spot in the upper atmosphere that can be optically imaged from the ground, putting the magnetic connection of the equatorial spacecraft's measurements into the context of the aurora. Multiple technical challenges that must be overcome for this mission concept are discussed: these include spacecraft charging, beam dynamics, beam stability, detection of the beam spot in the presence of aurora, and the safety of nearby spacecraft.

**Keywords:** aurora, space experiments, magnetosphere, ionosphere, electron beams

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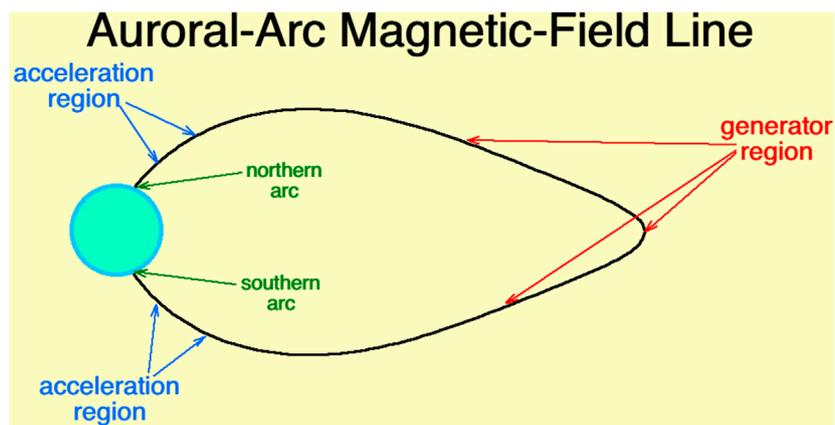
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## INTRODUCTION

One of the unsolved problems of magnetospheric physics is the cause of the various types of auroral forms (Lanchester, 2017; Denton, 2019). This is particularly the case for auroral arcs (Denton et al., 2016; Borovsky et al., 2020a), where an unknown generator mechanism in the equatorial and near-equatorial magnetosphere extracts power and current from the magnetosphere to drive an auroral arc that dissipates energy in the ionosphere and atmosphere. This is sketched in **Figure 1**. A large number of diverse generator mechanisms have been hypothesized (e.g., Borovsky, 1993; Haerendel, 2011; Haerendel, 2012; Borovsky et al., 2020b) but equatorial magnetospheric measurements have not been unambiguously connected to arcs in the ionosphere, preventing scientists from identifying the correct generator mechanisms. For quiescent auroral arcs, even the form of energy that is extracted from the magnetosphere (magnetic energy, ion thermal energy, electron thermal energy, flow kinetic energy, ...) is not known. The auroral community understands the near-Earth acceleration processes quite well (e.g., field-aligned potentials and Alfvén-wave electron acceleration) but does not understand the equatorial energy-conversion processes driving these near-Earth processes; nor does the community understand the origin of the Alfvénic energy. The aurora is a manifestation of complex processes operating in the distant magnetosphere; the desire to use optical images of the aurora as television-screen view of magnetospheric processes (e.g., Akasofu, 1965; Mende, 2016a;



**FIGURE 1** | A sketch of the equatorial magnetospheric spacecraft firing an electron beam into the northern loss cone where the beam spot can be located by a ground-based camera. Conjugate northern and southern auroral arcs are noted in the upper atmosphere, the near-Earth acceleration regions for auroral-arc electrons are indicated, and the equatorial and near-equatorial regions where the auroral-arc generator mechanisms operate are noted.

Mende, 2016b) is impeded by not knowing what processes act to create the various auroral forms.

The foundation of the auroral-cause problem is an inability to unambiguously connect equatorial magnetospheric measurements to the various auroral forms. Static magnetic models of the magnetosphere (e.g., Tsyganenko, 1989; Tsyganenko and Sitnov, 2007; Sitnov et al., 2008) are not sufficiently accurate to magnetically map ionospheric features into the equatorial electron plasma sheet, particularly into the high-Reynolds-number magnetotail (Borovsky et al., 1997; Voros et al., 2004; El-Alaoui et al., 2010; Stepanova et al., 2011; El-Alaoui et al., 2012). Tests of the accuracy of magnetic-field-line mapping with the standard magnetic-field models have found very large errors for the nightside magnetosphere (Thomsen et al., 1996; Weiss et al., 1997; Ober et al., 2000; Shevchenko et al., 2010; Nishimura et al., 2011), even in the quasi-dipolar regions. For quiescent, low-latitude (growth-phase) auroral arcs there are two schools of thought about the equatorial location of the source of the arc: one school assume that the arc is in the dipolar portion of the nightside magnetosphere (e.g., McIlwain, 1975; Meng et al., 1979; Kremser et al., 1988; Mauk and Meng, 1991; Pulkkinen et al., 1991; Lu et al., 2000; Motoba et al., 2015) and a second school has the source of the arc in the stretched magnetotail (e.g., Yahnin et al., 1997; Yahnin et al., 1999; Birn et al., 2004a; Birn et al., 2004b; Birn et al., 2012; Sergeev et al., 2012; Hsieh and Otto, 2014): our ability to map magnetic-field lines in the nightside magnetosphere is insufficient to determine which is the correct location.

Developing the technology to attain accurate “Magnetosphere-to-Ionosphere Field-Line-Tracing Technology” has been cited as an “instrument development need and emerging technology” necessary for the future of space science (National Research Council, 2012). For several decades a team of researchers centered around Los Alamos National Laboratory has worked to develop a viable spacecraft mission to unambiguously determine the magnetic connection between equatorial-magnetospheric measurements and optical auroral

observations (Borovsky et al., 1998; Borovsky, 2002; NASA, 2003; NASA, 2006; Delzanno et al., 2016; Borovsky and Delzanno, 2019; Borovsky et al., 2020c). That research team has consisted of auroral observers, magnetospheric instrument designers, optical physicists, ionospheric physicists, plasma physicists, spacecraft systems scientists, and two compact-accelerator design groups. That mission concept (and the technical challenges that it must overcome) is the focus of this brief report.

## THE MISSION CONCEPT: AN ELECTRON BEAM ILLUMINATING THE MAGNETIC CONNECTION BETWEEN AN EQUATORIAL MAGNETOSPHERIC SPACECRAFT AND THE ATMOSPHERE/IONOSPHERE

As sketched in **Figure 1**, the mission concept is for a magnetospheric spacecraft to carry an electron accelerator, to fire an electron beam along the magnetospheric magnetic field into the atmospheric loss cone, and with a ground-based camera to optically image the beam spot in the upper atmosphere. If that is accomplished then it is unambiguously known that a measurement taken by the magnetospheric spacecraft magnetically connects to the location in the ionosphere where the beam spot is imaged (Note that one can also approximately account for the eastward curvature-drift shift of the electron beam to more-accurately identify the magnetic location of the spacecraft.)

There are a number of challenges with this simple concept, and a good deal of research has been performed to overcome those difficulties: the technical challenges include spacecraft charging, beam aiming, beam dynamics and stability, and the detection of the beam spot in the presence of aurora. The major challenges and the associated mission-design tradeoffs are discussed in **Sections 3 and 4**.

Mission concepts have been examined that involve either (A) a single magnetospheric spacecraft making measurements and carrying an electron accelerator or (B) a swarm of measuring spacecraft with one member of the swarm carrying the electron accelerator. The spacecraft carrying the accelerator will also carry a power-storage system and a plasma contactor (for spacecraft-charging mitigation). The purpose of a swarm is to measure perpendicular-to- $\mathbf{B}$  gradients in the magnetosphere, which are important for diverting perpendicular magnetospheric currents into field-aligned currents, a critical part of the processes of driving of auroral arcs; the perpendicular gradients of interest are ion-pressure gradients, electron-pressure gradients, mass-density gradients, temperature gradients, flow shear, and gradients in the field strength, and the cross products of the various gradients are of interest (cf. eq. 12 of Strangeway 2012 or eq. 1 of Borovsky et al. 2020b). As analyzed in Borovsky et al. 2020c, the measurement requirements for quiescent auroral arcs in the equatorial magnetosphere appear in Table 1. For some theories of auroral arcs (e.g., Schindler and Birn, 2002; Birn et al., 2004a; Birn et al., 2012; Yang et al., 2013; Hsieh and Otto, 2014; Coroniti and Pritchett, 2014), Hall effects are important and so measuring both the perpendicular ion flow and the perpendicular electron flow is desirable; this can be accomplished by measuring both the ion flow and the electric field.

Two mission concepts have been considered. The first, Magnetosphere-Ionosphere Observatory (MIO) (Borovsky et al., 1998; Borovsky, 2002) consists of a tight (100's of km) swarm of spacecraft in the equator at geosynchronous orbit ( $6.6 R_E$  where  $R_E$  is the radius of the Earth), with a single ground-based observatory in the vicinity of the swarm's magnetic footpoints. In Figure 2 the location at 100 km altitude of the magnetic footpoint of a spacecraft in the geographic equator at geosynchronous orbit is estimated using the T89 (Tsyganenko, 1989) and the IGRF (Maus et al., 2005) magnetic-field models. The spacecraft is located in the "Alaska sector" of geosynchronous orbit where the geographic and geomagnetic equators are close to each other. The "observatory" in the figure is located at Eagle, Alaska ( $64^{\circ}47' N, 121^{\circ}12' W$ ). The red circles in Figure 2 are the angle from zenith where the 100 km altitude is seen by the observatory. In Figure 3 a similar plot is made for a spacecraft in the geographic equator at geosynchronous orbit in the "Scandinavian sector". The ground-based observatory would have at least one camera dedicated to beam spot imaging, although, owing to uncertainty in the estimation of the magnetic-footpoint location using magnetic-field models, a network of beamspotting cameras around the observatory location will probably be needed. The observatory would have cameras for auroral imaging and other instrumentation for ionospheric physics. In the MIO mission concept a ground-based radar could be used to help locate the beam spot (e.g., Izhovkina et al., 1980; Uspensky et al., 1980; Zhulin et al., 1980; Marshall et al., 2014); additionally the radar could be used for physics studies with the electron beam as an element of upper-atmosphere experiments. Other instrumentation at the observatory could be ionosondes, an

TABLE 1 | Magnetospheric measurement requirements for quiescent arcs.

Measurement quantity	Typical value	Desired accuracy
Proton number density	$1 \text{ cm}^{-3}$	$0.1 \text{ cm}^{-3}$
Proton temperature	10 keV	1 keV
Proton pressure	1.6 nPa	0.16 nPa
Proton pressure anisotropy	0.16 nPa	–
Electron number density	$1 \text{ cm}^{-3}$	$0.1 \text{ cm}^{-3}$
Electron temperature	2 keV	200 eV
Electron pressure	0.32 nPa	0.03 nPa
Electron pressure anisotropy	0.03 nPa	–
Proton flow along arc	150 km/s	15 km/s
Electron flow along arc	150 km/s	15 km/s
Proton or electron flow across arc	6.6 km/s	1.5 km/s
Magnetic-field direction vector	–	$0.5^{\circ}$
Magnetic-field strength	80 nT	1 nT

ionospheric heater, a wave transmitter, and a magnetometer network. An important aspect of the MIO mission concept is the ability to concentrate ground-based infrastructure at a single location.

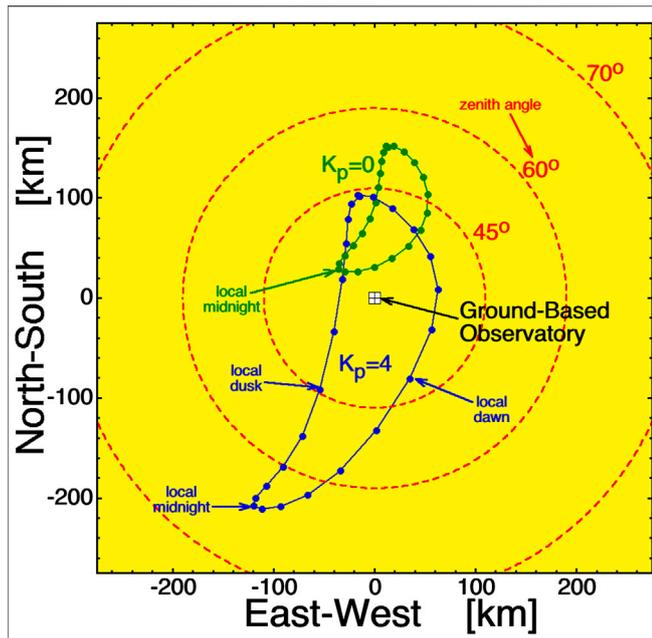
The second mission concept, called connections (Borovsky et al., 2020c), has a swarm of spacecraft in an eccentric orbit and takes advantage of the Canadian TREx (Transition Region Explorer) (Spanswick et al., 2018) network of auroral cameras. Orbits are chosen with periods of 24 h so that the magnetic footpoint of the swarm wanders over Western Canada with a 24-h period. A  $5 R_E$  by  $8 R_E$  orbit was studied for its desirable footpoint locations (Borovsky et al., 2020c); note with an 8- $R_E$  apogee, the spacecraft swarm can magnetically map further downtail than  $8 R_E$  owing to orbital inclination and dipole tilt. The approximate magnetic-footpoint locations for the  $5 R_E$  by  $8 R_E$  24-h orbit are shown as the blue curve in Figure 4 and the approximate footpoint locations of a  $4 R_E$  by  $8 R_E$  24-h orbit are also shown as the yellow curve in Figure 4. In Figure 4 the field-of-views of the individual (Transition Region Explorer) TREx cameras are indicated as the red circles. An eccentric 24 h orbit can better sample the stretched-magnetotail portion of the nightside magnetosphere than can a circular geosynchronous orbit.

## CHALLENGES TO OVERCOME

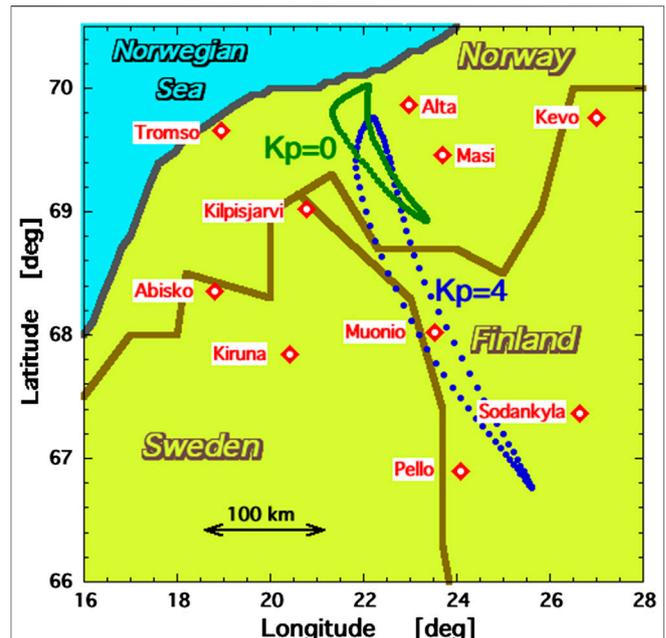
Four challenges that must be overcome for this magnetospheric-electron-beam mission concept are discussed in this section.

### Spacecraft Charging

The most critical issue for operating a high-power electron beam from an ungrounded spacecraft in the tenuous magnetospheric plasma is spacecraft charging. The beam must deposit sufficient power ( $\sim 5 \text{ kW}$ ) in the upper atmosphere to be seen in the presence of ongoing aurora: if the beam energy is 50 keV then a beam current of 100 mA is required and if the beam energy is 1 MeV then a beam current of 5 mA is required. Firing the beam will result in a fraction of a Coulomb of negative being removed from the spacecraft in a time on the order of 1 s. A substantial computer-simulation-based research effort (Delzanno et al.,



**FIGURE 2** | Looking down onto Alaska, the approximate location each hour of the day in Winter of the magnetic footpoint at 100 km altitude of a spacecraft in the geosynchronous orbit geographic equator is plotted. The green points are for  $K_p = 0$  and the blue points are for  $K_p = 4$ . The observatory is located at Eagle, Alaska. The red circles are the zenith angle of the 100 km altitude as seen from the ground-based observatory.



**FIGURE 3** | Looking down onto Scandinavia, the approximate location each hour of the day in Winter of the magnetic footpoint at 100 km altitude of a spacecraft in the geosynchronous orbit geographic equator is plotted. The green points are for  $K_p = 0$  and the blue points are for  $K_p = 4$ .

2015a; Delzanno et al., 2015b; Delzanno et al., 2016; Lucco Castello et al., 2018), supported by laboratory experiments (Miars et al., 2020), has demonstrated that the operation of a plasma contactor releasing a high-density charge-neutral plasma plume before and during a beam firing can greatly mitigate the charging of the spacecraft during the beam operation. Contrary to prior discussion of an emitted plasma plume acting to collect charge from the ambient plasma (e.g., Hastings and Blandino, 1989; Gerber et al., 1990; Williams and Wilbur, 1990; Davis et al., 1991), the research effort demonstrated that the surface of the plasma plume acts as an ion emitter, producing an ion current equal to the current of the electron beam.

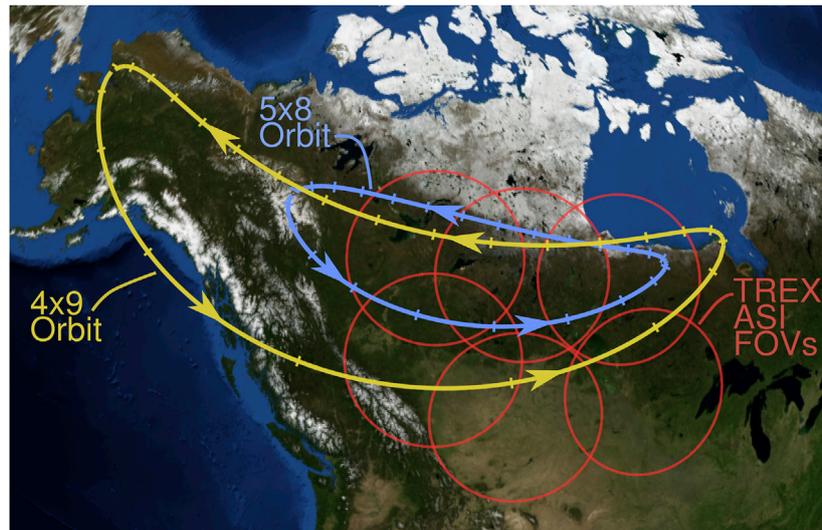
### Getting the Beam to the Atmosphere

Getting the electron beam from the spacecraft in the magnetospheric equator to the atmosphere involves aiming the beam into the atmospheric loss cone, fitting the beam within the loss cone, ensuring that the propagating beam is stable, and ensuring that the propagating beam electrons are not scattered by magnetospheric plasma waves.

Assuming that the magnetic-field strength in the auroral upper atmosphere is  $\sim 0.5$  Gauss, the radius of the atmospheric loss cone is  $2.5^\circ$  if the spacecraft is in a 100 nT field (e.g., in geosynchronous orbit) and the radius is  $1.1^\circ$  if the spacecraft is in a 20 nT field (e.g., in the stretched magnetotail). Knowledge of the direction of the ambient magnetic field to an accuracy of about  $0.5^\circ$  is needed, and an ability to aim the beam with an accuracy of about  $0.5^\circ$  is also needed. A complication to the aiming into the

loss cone occurs if the electron beam is very energetic: finite-gyroradii effects shift the direction of the loss cone eastward (for electrons) from the local magnetic-field direction (Il'ina et al., 1993; Mozer, 1966; Porazik et al., 2014; Powis et al., 2019; Willard et al., 2019; Borovsky et al., 2020c). For a dipole magnetic field the magnitude of this eastward angular shift is easily predictable (Mozer, 1966; Borovsky et al., 2020c), but for non-dipolar magnetic fields this shift is not predictable and a space experiment would need to determine the loss-cone shift by repeatedly test firing the electron beam with differing amounts of eastward shift while ground cameras work to detect the beam spot. If onboard energy-storage resources are limited, this would not be desirable, and if the optical beam-spot-location image analysis is not instantaneous, this trial-and-error methodology cannot be implemented.

After the electron beam is emitted from the accelerator and as it travels along the Earth's magnetic field, the nonzero net negative space charge of the beam acts to repulsively accelerate beam electrons transverse to the magnetic field; this transverse acceleration results in a spread of pitch angles of the beam electrons, turning a narrowly focused beam into a "shotgun" (cf. Appendix B of Borovsky, 2002). The space charge per unit length of the beam  $Q/L$  is given by  $Q/L = I_{\text{beam}}/v_{\text{beam}}$ , where  $I_{\text{beam}}$  is the current of the beam and  $v_{\text{beam}}$  is the speed of the beam. The beam power  $P_{\text{beam}}$  is  $P_{\text{beam}} = I_{\text{beam}}V_{\text{beam}}$ , where  $V_{\text{beam}}$  is the beam voltage (accelerator energy). The speed of the beam increases with the beam voltage. For the same amount of beam power, a high-voltage beam has less current and higher speed, hence it has much less charge per unit length  $Q/L$ , and the space-



**FIGURE 4 |** The estimated magnetic footpoints over Canada and the (Transition Region Explorer) TREx fields of view of two elliptical orbits: a  $5 R_E \times 8 R_E$  24 hr-period orbit (blue) and a  $4 R_E \times 9 R_E$  24 h orbit (yellow). The arrows on the footpoint curves are the temporal direction of the movement of the footpoint relative to the ground. The red circles are the fields-of-view at 100 km altitude of the present (Transition Region Explorer) TREx cameras. The low-latitude portions of the footpoint curves correspond to spacecraft perigee's and the hash marks on the footpoint curves are at 1 h intervals in the 24 h orbits.

charge transverse spreading of the beam is much less. Figure 2 of Borovsky et al., (2020c) looks at the maximum power of a beam that will stay within the loss cone as a function of the beam voltage. Greater beam voltage is very advantageous from the beam-spreading point of view.

Calculating the stability of the electron beam propagating through the ambient magnetospheric plasma is an ongoing area of research. Fortunately, the powerful electrostatic two-stream instabilities are greatly weakened by the fact that the beam has a small cylindrical cross section (Galvez and Borovsky, 1988). Relativistic electron beams have been calculated to be stable to electromagnetic hose and filamentation instabilities (Gilchrist et al., 2001; Neubert and Gilchrist, 2002; Neubert and Gilchrist, 2004). Experimentally, similar beams have been detected after long-distance propagation through the magnetosphere. Beams with energies of up to 40 keV were propagated long distances through the magnetosphere in the Echo series of experiments (Hallinan et al., 1990; Winckler, 1992) and electron beams of 27 keV, 0.5 Amp and 15 keV, 0.5 Amp on the two ARAKS experiments were propagated  $8.2 R_E$  through the magnetosphere without disruption (Pellat and Sagdeev, 1980; Lavergnat, 1982).

Finally, an issue of ongoing research is estimating the amount of pitch-angle scattering that beam electrons will undergo from the action of ambient magnetospheric plasma waves when the spacecraft is in various locations of the magnetosphere during different levels of geomagnetic activity. The degree of pitch-angle scattering will vary significantly with the beam voltage. At higher (relativistic) beam voltages electromagnetic ion-cyclotron (EMIC) waves and whistler-mode chorus are of concern, as is field-line curvature acting to scatter the beam. At lower beam voltages whistler-mode chorus and electromagnetic electron-

cyclotron waves are of concern. Section 6.3 of Borovsky et al. (2020c) provides some preliminary estimates of the amount of pitch-angle scattering of beam electrons in the nightside magnetosphere: those estimates are favorable for the beam surviving from the equator to the atmosphere. Note also that (electromagnetic ion-cyclotron) EMIC waves are prevalent in the noon and afternoon sectors and not prevalent in the nightside auroral zone (Clausen et al., 2011; Usanova et al., 2012).

### Detecting the Beam Spot

To produce a beam spot that is optically detectable from the ground in the presence of active aurora, a beam power of 5 kW or more needs to be deposited in the atmosphere. Each 1 kW of beam power results in about 1.1 W of  $4278 \text{ \AA}$  emission [Bryant et al., 1970] and about 3 W of  $3914 \text{ \AA}$  emission. The spectral lines emitted by the beam spot will be the same spectral lines as emitted by the electron aurora. The beam spot will be cylindrical, 10's of m in diameter across the magnetic field and  $\sim 10$  km long along the magnetic field; this will produce a multi-pixel streak in the camera images. The technology of optically detecting the illuminated footpoint from the ground has been verified: electron beams with less than 5 kW of power have been optically detected from the ground after they have propagated through the magnetosphere into the upper atmosphere. Two examples are the detection of a 3.4 kW beam by Davis et al. (1980) during the NASA 12.18 NE beam experiment and the detection of a 2.4 kW beam in the Echo-4 experiment by Hallinan et al. (1990) using a ground-based image-orthicon television. In the Echo-4 experiment, the beamspot was imaged after the beam had propagated twice through the magnetosphere at  $L = 6.5$ . To identify the beam spot in the presence of aurora, an on-off temporal blink pattern of beam firings must be used along

with temporal processing of the camera images to locate the blinking streak. If the mission has a “scientist in the loop” commanding the firing of the beam as the beam spot crosses critical auroral features, the image processing must be automatic and prompt.

For beams of 10’s of keV the beam spot altitude is around 90–100 km; for beams of 1 MeV the beam spot altitude is around 60 km. If the beam has a narrow spread of pitch angles, the beam spot altitude can be raised by aiming the beam away from the center of the loss cone. At 60 km there is some collisional quenching of the 3914 and 4278 Å prompt emission bands (Marshall et al., 2014; Borovsky et al., 2020c), so more beam power is required to image the beam spot. Unless it is overhead of the camera, a beam spot at 60 km altitude will also suffer more extinction from Rayleigh scattering (Penndorf, 1957; International Telephone and Telegraph Corporation, 1977), again requiring more beam power.

### Safety of Other Magnetospheric Spacecraft

Since there are other spacecraft in orbit around the Earth, an important aspect to consider is whether the electron beam emitted could intercept another spacecraft and induce catastrophic charging on it. After the beam is emitted, it expands and contracts periodically with time due to space-charge and Lorentz forces as well as due to spreading in pitch angle and energy due to the electron accelerator design (Borovsky, 2002; Powis et al., 2019). As a result the beam current density will change significantly along the beam path. Any mission design will need a beam-safety plan. To estimate the beam flux to another spacecraft, the beam dynamics must be modeled and beam-connection dwell times with other spacecraft must be calculated using orbital considerations.

From preliminary calculations of the dynamics of a 1 MeV 10 mA beam in the dipolar magnetosphere with a criterion that the current flux to a second spacecraft must be less than  $10^{-6}$  A/m<sup>2</sup> yields a “current safety distance” of 0.5 R<sub>E</sub> along the Earth’s magnetic field from the accelerator to another spacecraft. In the tenuous plasma of the magnetosphere an electron beam flux of  $10^{-6}$  A/m<sup>2</sup> would only induce ~1 kV of spacecraft charging to the second spacecraft.

## MAJOR TRADEOFFS FOR A MISSION

There are many tradeoffs that must be made in designing a mission. A major tradeoff is whether to have a single-observatory geosynchronous mission or an elliptic orbit distributed-camera-network mission (cf. Sect. 2). Two other major tradeoffs are discussed below.

### Relativistic vs. Nonrelativistic Electron Beams

The accelerator technology for relativistic (~MeV) vs. nonrelativistic (10’s-of-keV) beams differs: 10’s-of-keV beams can be produced with direct-current electron guns that accelerate the electrons through a static potential drop whereas

MeV beams must be produced with a radio-frequency electron accelerator that accelerates the electrons with a propagating wavefront. Direct-current electron guns with 10’s of keV energies and 10’s of kW powers have been flown in space numerous times (Winckler et al., 1975; O’Neil et al., 1978; Rappaport et al., 1993; Prech et al., 1995; McNutt et al., 1995; Prech et al., 2018), while a radio-frequency accelerator has only been flown once (a 1-MeV H<sup>-</sup> beam) (Pongratz, 2018). Designs for compact space-based relativistic-electron accelerators are underway (Lewellen et al., 2019) and spaceflight tests of the accelerator concepts are planned (Reeves et al., 2020).

The advantages of a relativistic electron accelerator over a 10’s-of-keV electron gun are 1) lower beam space charge for the same beam power, resulting in a beam with less angular spread to more-easily fit into the loss cone, and 2) lower total charge removed from the spacecraft in a beam pulse, requiring less-stringent spacecraft-charging mitigation. For beam energies below a few 10’s of keV, the space charge of the beam drastically limits the amount of beam power that can be delivered into the loss cone.

The disadvantages of a relativistic accelerator compared with a 10’s-of-keV electron gun are 1) a lower-altitude beam spot subject to quenching, 2) for a non-dipole magnetic field the location of the loss cone is not known, 3) more difficulty in steering the beam mechanically or electrostatically, 4) there can be accelerator-thermal issues that de-tune the radio-frequency cavities after several beam firings, and 5) there are more-critical safety considerations for other spacecraft. For beams with energies above about 1 MeV, the loss-cone shift becomes severe.

The relativistic vs. nonrelativistic beams will also have tradeoffs concerning beam stability and scattering and the accelerator and gun will have tradeoffs concerning the power-conversion efficiency and mass of the required energy-storage system. For any mission with a powerful electron beam, energy storage will require substantial spacecraft mass.

### Spinning vs. 3-Axis Stabilized Spacecraft

A spinning spacecraft is better, in general, for measuring plasma properties and fields. Spinning has the distinct advantage that offsets in the magnetometer can be corrected, resulting in a more-accurate determination of the magnetic field direction for beam aiming. If a relativistic accelerator is used, which will have a length of 1 m or so, mechanically steering the accelerator into the moving loss cone seen by a spinning spacecraft will be challenging. At relativistic energies, electrostatic steering of the electron beam after it exits the accelerator is very limited in angle.

A 3-axis stabilized spacecraft makes it easier to direct the beam into the loss cone, particularly if a long (0.5-s or 1-s) beam pulse is used. A de-spun platform on a spinning spacecraft is another option.

Note that if an electron drift instrument (Torbert et al., 2016) is used to measure the electric field, that instrument works in concert with a magnetometer and magnetometer offsets can be detected and corrected, even on a non-spinning spacecraft.

## OTHER SCIENCE

Most of the work on the development of this magnetospheric-electron-beam mission concept has been motivated by the desire to understand the generator mechanisms of low-latitude quiescent auroral arcs (Borovsky, 2002; Delzanno et al., 2016; Borovsky et al., 2020c). However, such a beam experiment could be used to explore a wide variety of scientific problems.

The magnetospheric causes of various other types of aurora could be investigated if there is appropriate instrumentation on the magnetospheric accelerator spacecraft or the swarm spacecraft. Discerning the causes of other aurora may require wave measurements, cold-ion and cold-electron measurements, and particle-anisotropy measurements: examples of this would be the diffuse and pulsating aurora. Other auroral forms include undulations of the equatorward auroral boundary, omega bands, torches, black aurora, and patches. Investigating high-latitude auroral forms such as streamers or high-latitude Alfvénic arcs will require a spacecraft orbit that is eccentric.

The mapping of boundaries and regions between the magnetosphere and the ionosphere could be unambiguously performed with a magnetospheric-electron-beam spacecraft, provided it is instrumented to identify those boundaries and regions. Of interest are the mapping to the ionosphere of a) the inner edge of the electron plasma sheet, b) the remnant layer, c) the plasmopause, d) detached plasmasphere regions, e) ion-isotropy boundaries, f) substorm-injection boundaries, and g) the Earthward edge of the cross-tail current sheet. The magnetospheric boundaries (a)–(f) are clearly seen at the geosynchronous-orbit equator: the magnetic field at geosynchronous orbit often exhibits a stretched-tail morphology at local midnight (cf. Figure 11 of Borovsky and Denton (2010)) but boundary (g) is ambiguous to detect in the magnetosphere. Conversely, the mapping of ionospheric troughs out into the magnetosphere is of interest.

The magnetic connections between magnetospheric and ionospheric processes such as SAPS, SAID, STEVE, convection reversals, and bursty bulk flows could be determined with certainty. An eccentric orbit provides more-regular access to these various phenomena, in particular to bursty bulk flows.

Magnetosphere-ionosphere coupling can be studied by comparing temporal onsets of convection in the magnetosphere (via spacecraft flow measurements) with temporal onsets of ionospheric convection (measured, for example, by the SuperDARN radar network (Greenwald et al., 1995; Baker et al., 2011; Bristow et al., 2016)) and could answer questions about when and where the magnetosphere drives ionospheric convection and when and where the ionosphere drives magnetospheric convection.

Finally, there is an extensive literature describing how energetic electron beams could be used to study mesospheric chemistry (Neubert et al., 1990; Marshall et al., 2019), atmospheric electricity (Banks et al., 1990; Neubert et al., 1990; Neubert and Gilchrist, 2004; Marshall et al., 2019; Sanchez et al., 2019; Borovsky et al., 2020c),

atmospheric electron-attachment physics and electrical conductivity (Banks et al., 1990; Neubert et al., 1996; Neubert and Gilchrist, 2004; Borovsky, 2017), and plasma-wave generation (Carlsten et al., 2018; Delzanno and Roytershteyn, 2019; Reeves et al., 2020).

## DATA AVAILABILITY STATEMENT

The raw data supporting the conclusions of this article will be made available by the authors, without undue reservation, to any qualified researcher.

## AUTHOR CONTRIBUTIONS

All authors performed research on this mission concept and contributed information for the manuscript.

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**Conflict of Interest:** The authors declare that the research was conducted in the absence of any commercial or financial relationships that could be construed as a potential conflict of interest.

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